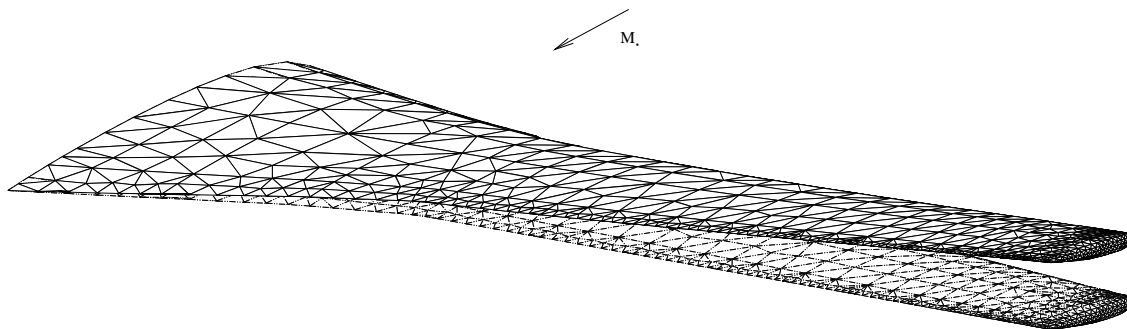


(c) Supercritical wing flow ($M_\infty = 0.85$, $\alpha = 2.5^\circ$, altitude = 35,000 ft).



(d) Supersonic wing flow ($M_\infty = 1.2$, $\alpha = 2.5^\circ$, altitude = 35,000 ft).

Figure 6.24: Concluded.